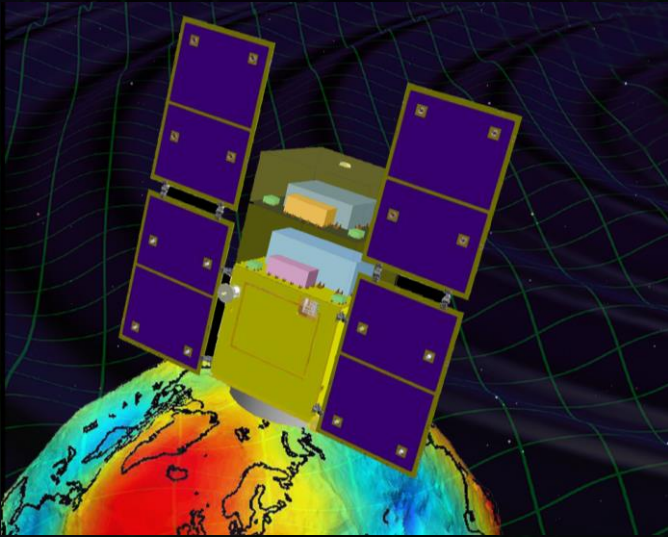


DECIGO and DECIGO Pathfinder



Masaki Ando

(Dept. of Physics, Univ. of Tokyo /
National Astronomical Observatory Japan)

DECIGO Members



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(On February 28th, 2014)

- **DECIGO**
- **DECIGO Pathfinder**

DECIGO

Space GW Antenna DECIGO



DECIGO (DECI-hertz interferometer
Gravitational wave Observatory)

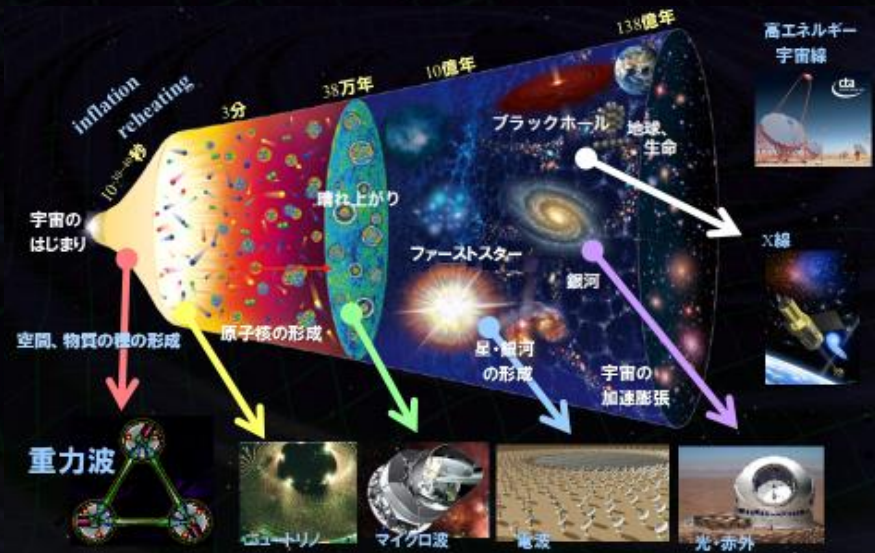
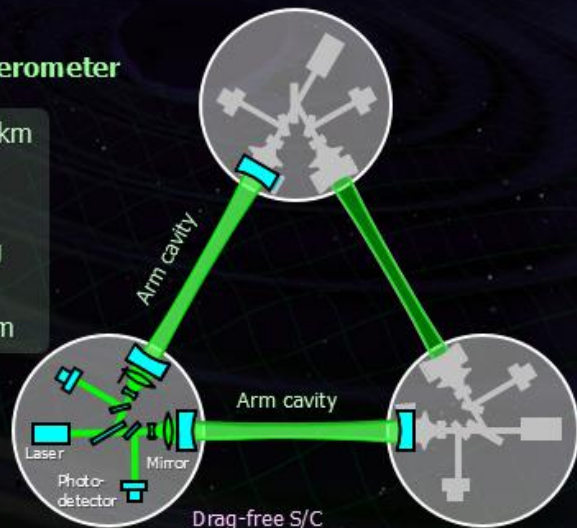
Purpose: To Obtain Cosmological Knowledge.

Direct observation of the origin of space-time
and matter in Big-bang Universe.

Interferometer Unit:
Differential FP interferometer

Arm length: 1000 km
Finesse: 10
Mirror diameter: 1 m
Mirror mass: 100 kg
Laser power: 10 W
Laser wavelength: 532 nm

S/C: drag free
3 interferometers



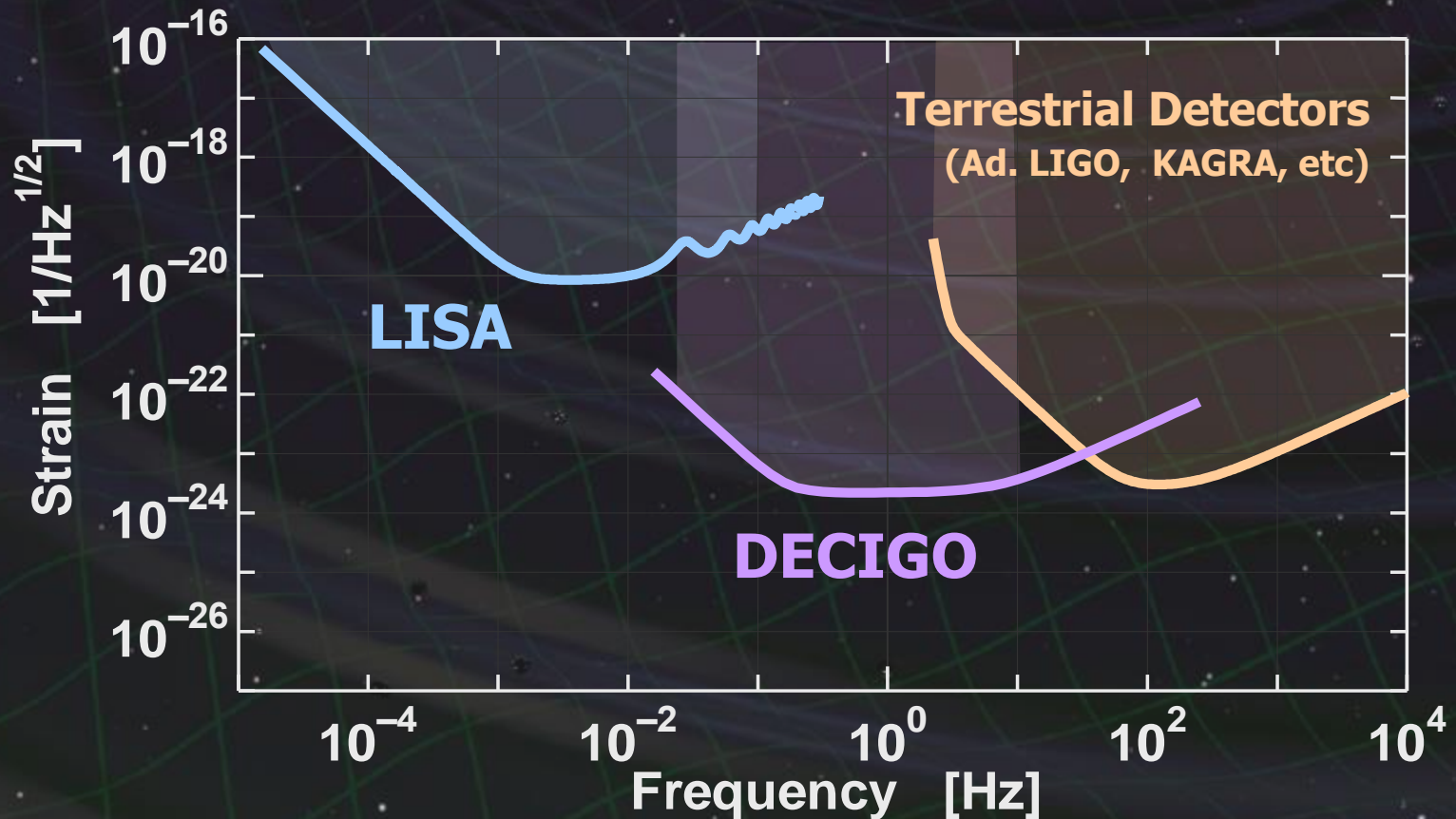
背景画: 福井康雄監修「宇宙史を物理学で読み解く
-素粒子から物質・生命まで」(名古屋大学出版会)より

DECIGO (Deci-hertz interferometer Gravitational wave Observatory)

Space GW antenna (~2027)
Obs. band around 0.1 Hz



'Bridge' the obs. gap between
LISA and Terrestrial detectors

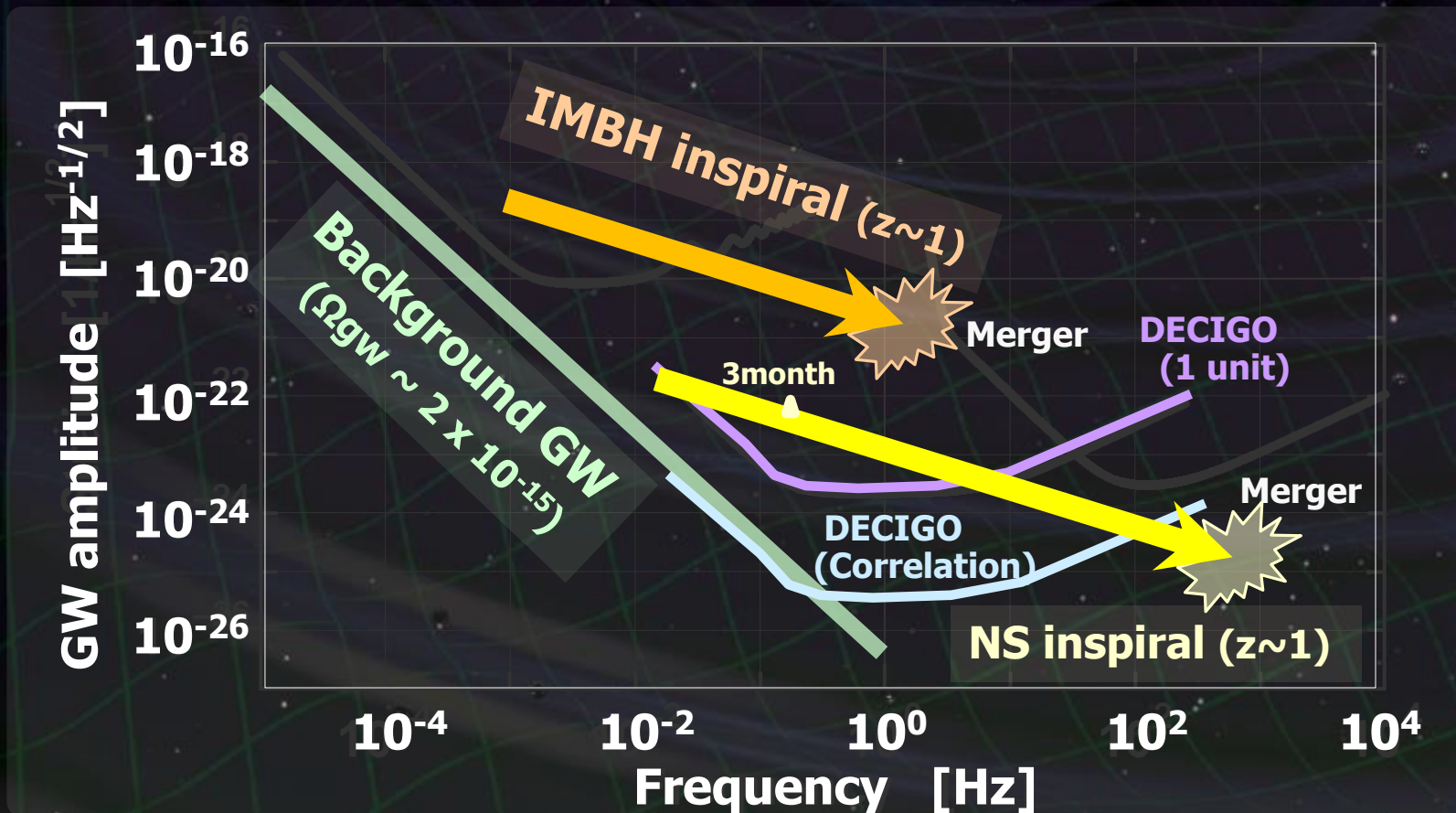


Targets and Science

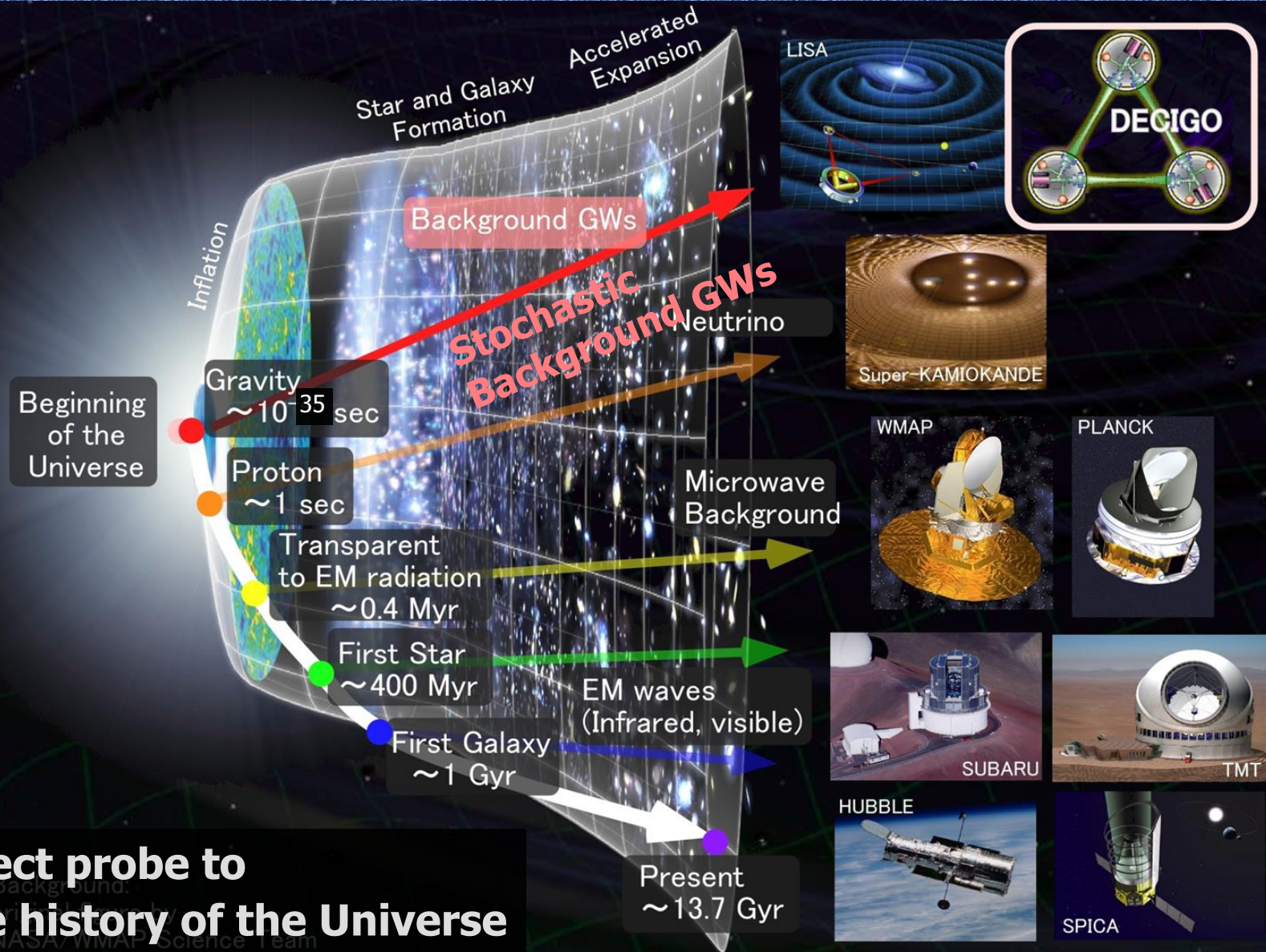
IMBH binary inspiral
NS binary inspiral
Stochastic background



Galaxy formation (Massive BH)
Cosmology (Inflation, Dark energy)
Fundamental physics



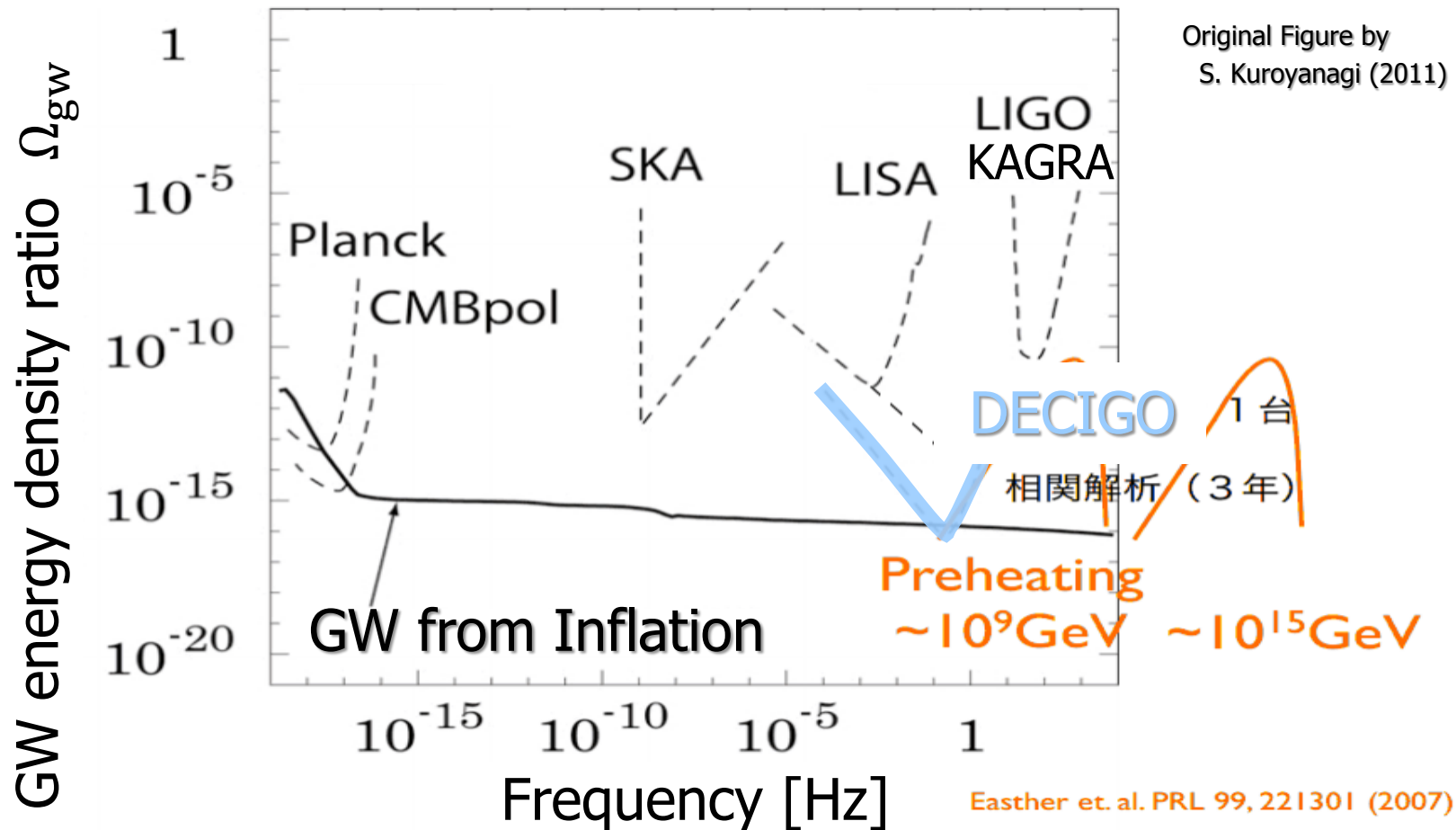
Characterization of inflation



Direct probe to the history of the Universe

Primordial GW

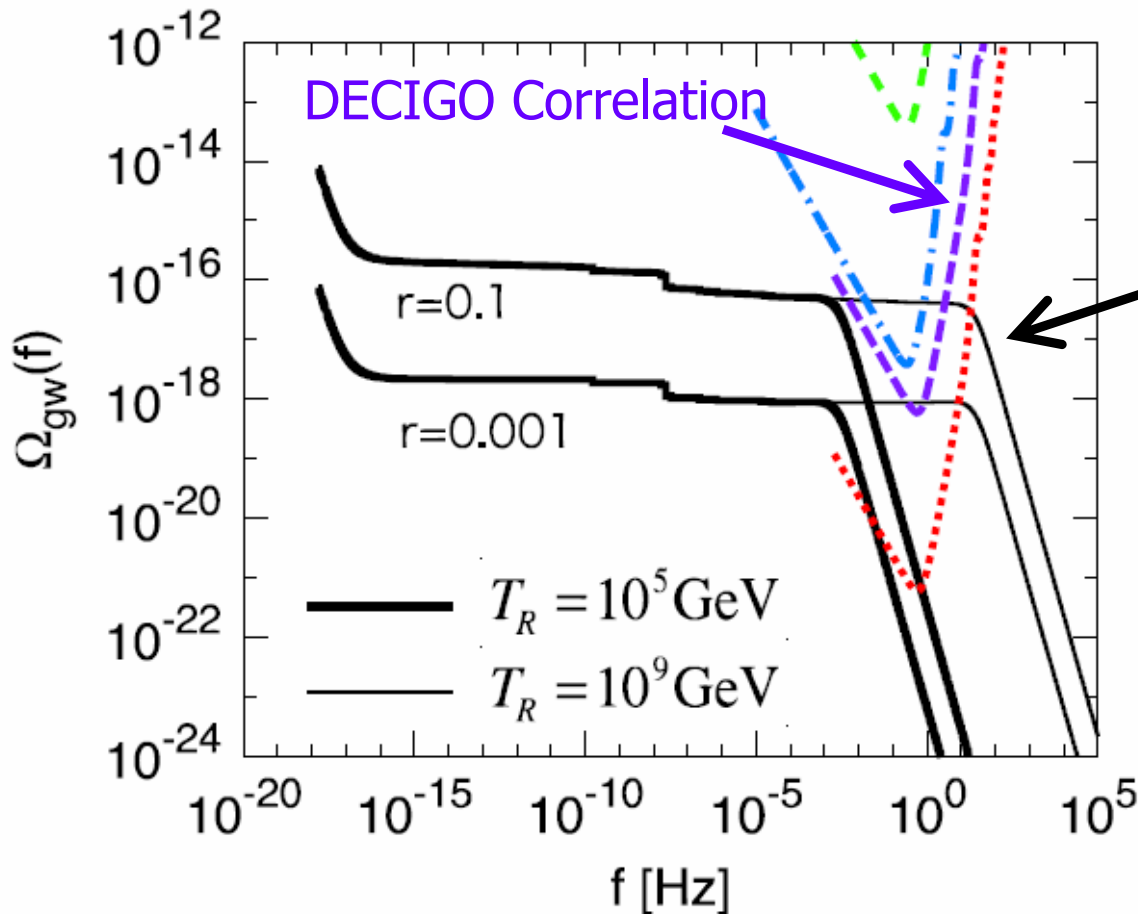
Earlier universe \rightarrow Smaller horizon scale \rightarrow High GW freq.



GW from Inflation

Energy density \propto Tensor-Scalar Ratio (r).

Power spectrum : Evolution history of the Universe.



- Spectrum Power.
→ Energy scale of inflation
- Cut-off freq.
→ Energy scale of Reheating
'Age of Big-bang'.

Nakayama+,
Journal of Cosmology
and Astroparticle Physics
06 (2008) 020.

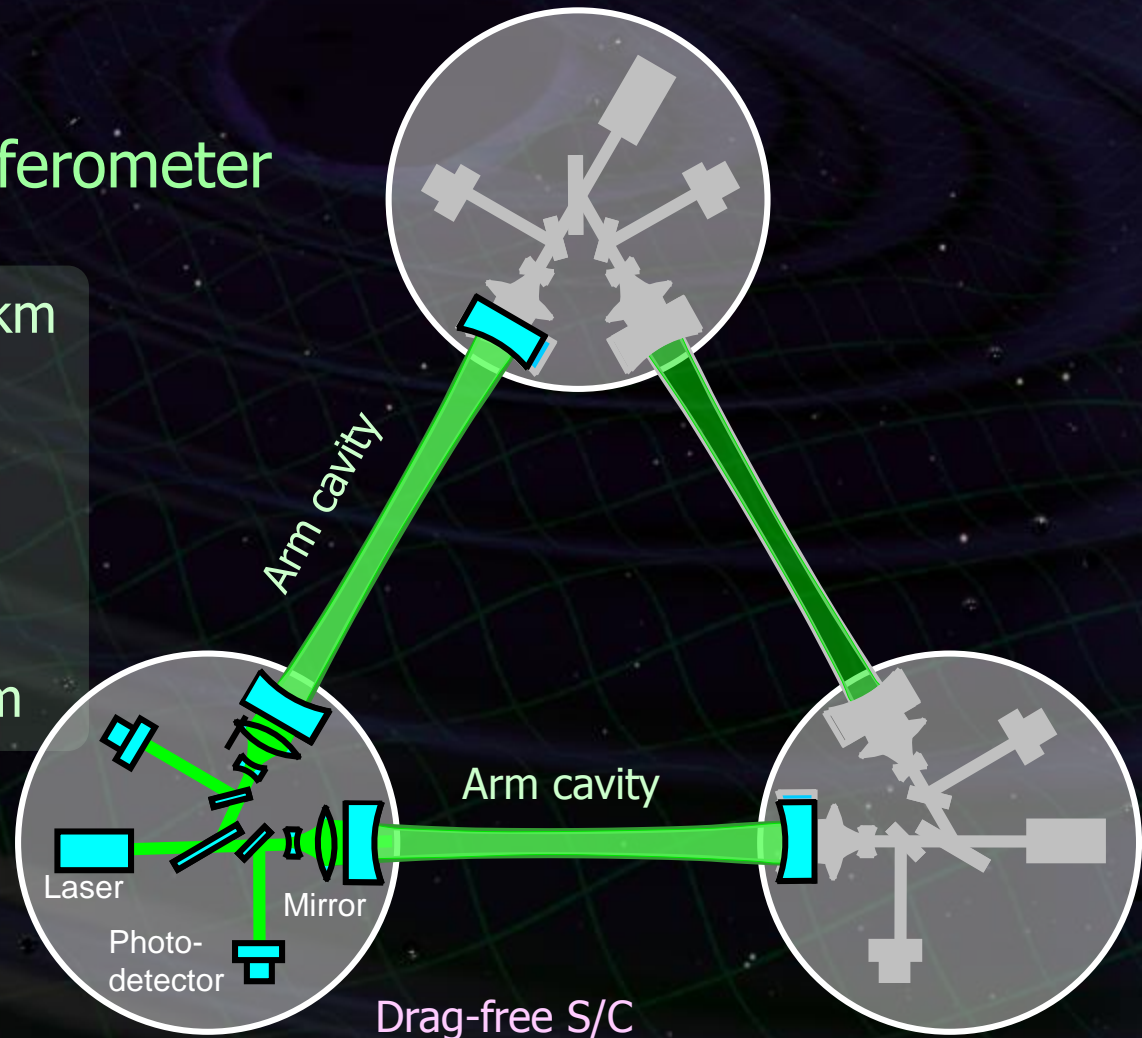
Pre-Conceptual Design

Interferometer Unit:

Differential FP interferometer

Arm length:	1000 km
Finesse:	10
Mirror diameter:	1 m
Mirror mass:	100 kg
Laser power:	10 W
Laser wavelength:	532 nm

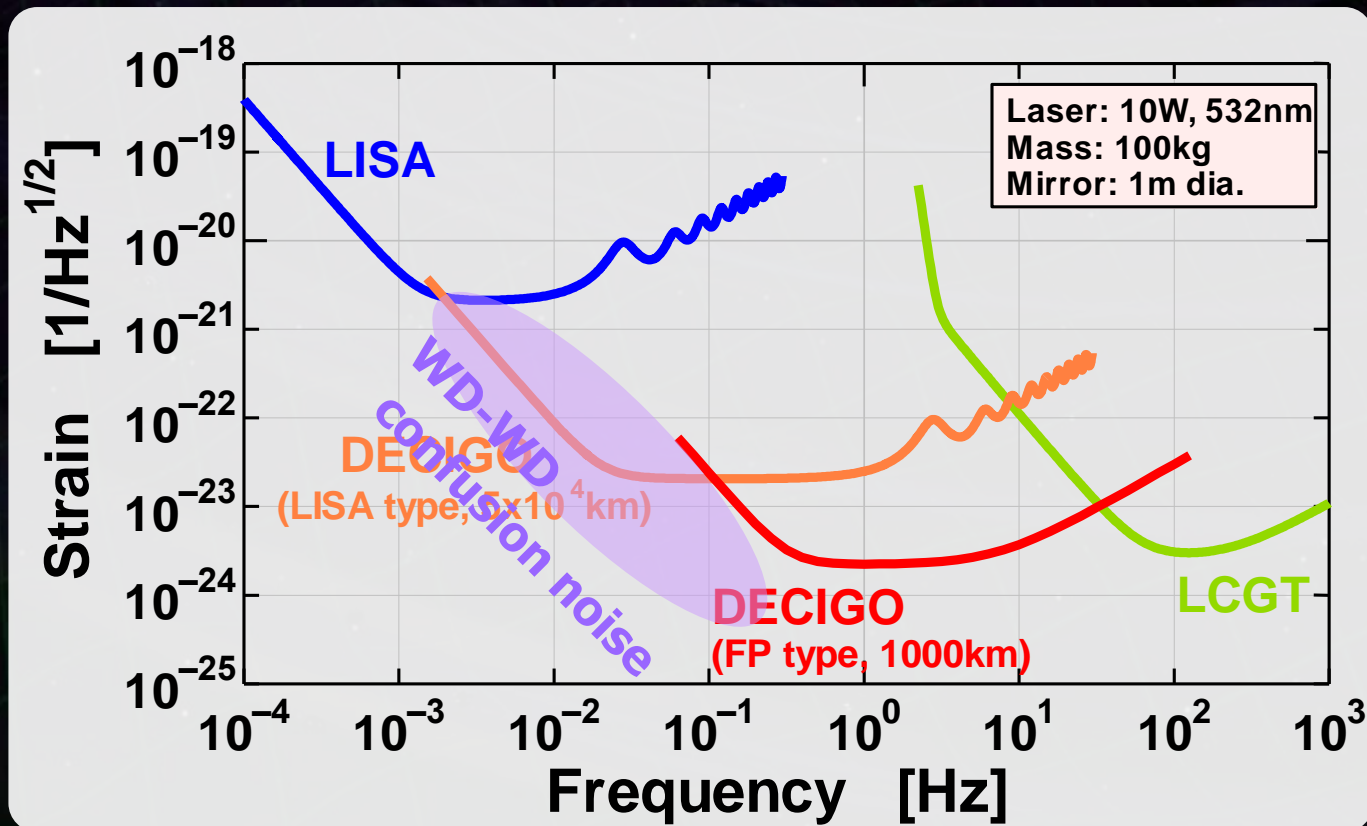
S/C: drag free
3 interferometers



Transponder type vs Direct-reflection type

Compare : Sensitivity curves and Expected Sciences

⇒ Decisive factor: Binary confusion noise



Arm length

Cavity arm length : Limited by diffraction loss

Effective reflectivity ($TEM_{00} \rightarrow TEM_{00}$)

Laser wavelength : 532nm

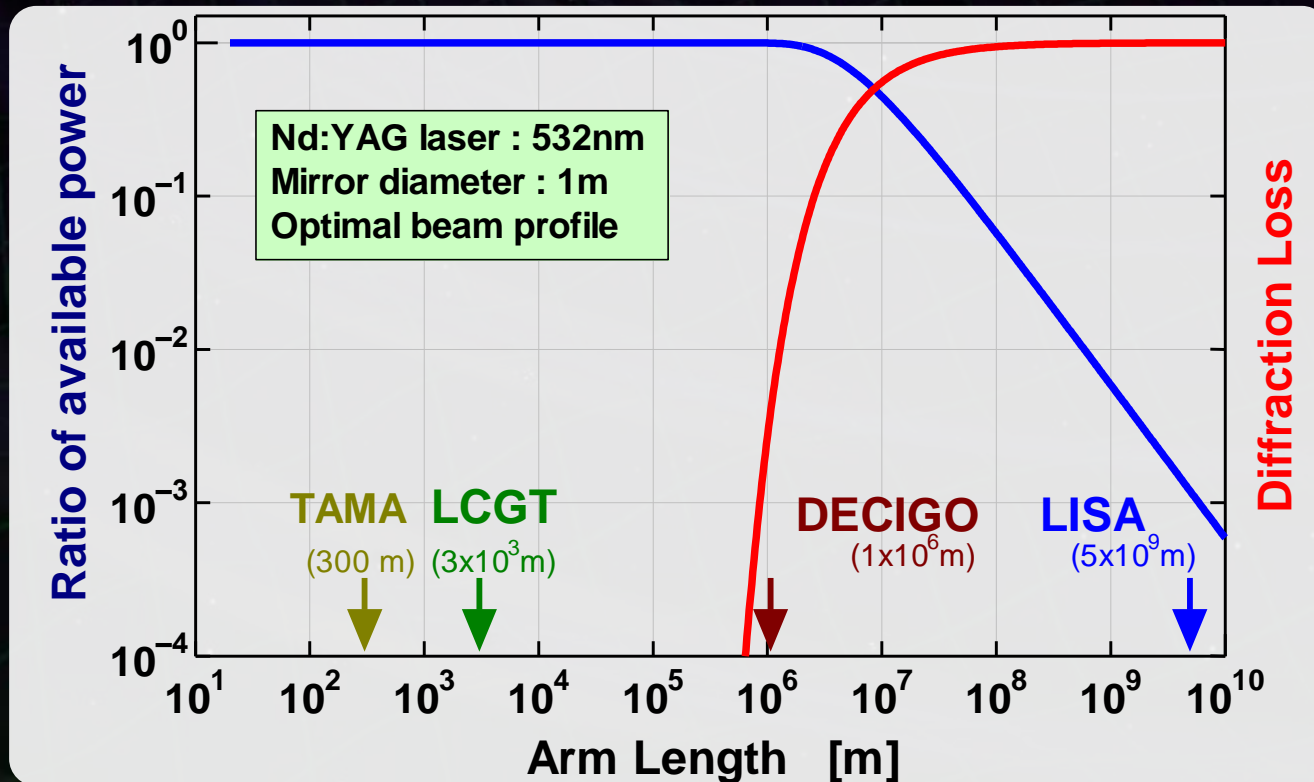
Mirror diameter: 1m

Optimal beam size



1000 km

is almost max.



Cavity and S/C control

Cavity length change

PDH error signal \rightarrow Mirror position (+Laser freq.)

Relative motion between mirror and S/C

Local sensor \rightarrow S/C thruster

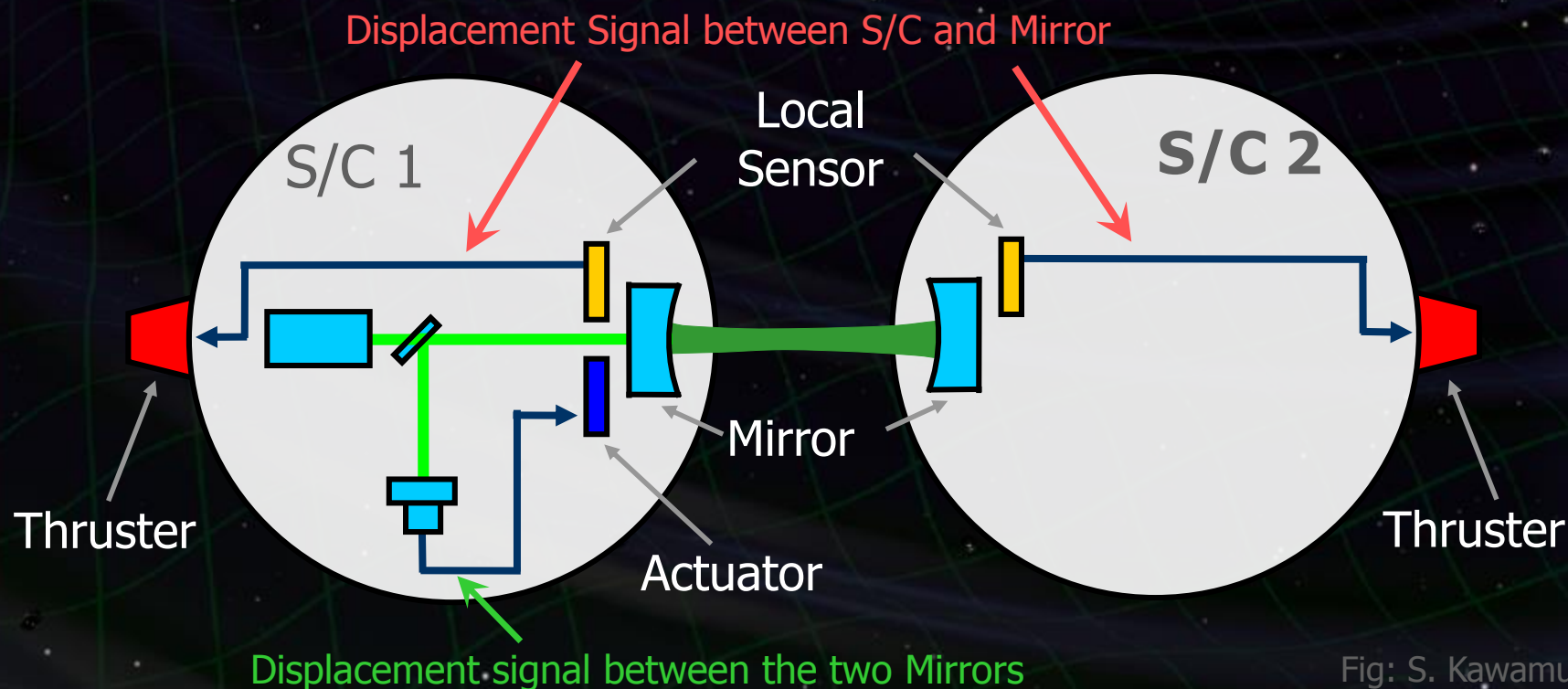


Fig: S. Kawamura

Displacement Noise

Shot noise $3 \times 10^{-18} \text{ m/Hz}^{1/2}$ (0.1 Hz)

⇒ x 10 of KAGRA in phase noise

Other noises should be well below the shot noise

Laser freq. noise: $1 \text{ Hz/Hz}^{1/2}$ (1Hz)

Stab. Gain 10^5 , CMRR 10^5

Acceleration Noise

Force noise $4 \times 10^{-17} \text{ N/Hz}^{1/2}$ (0.1 Hz)

⇒ x 1/50 of LISA

External force sources

Fluctuation of magnetic field, electric field,
gravitational field, temperature, pressure, etc.

Foreground Cleaning

DECIGO obs. band: free from WD binary foreground
→ Open for cosmological observation

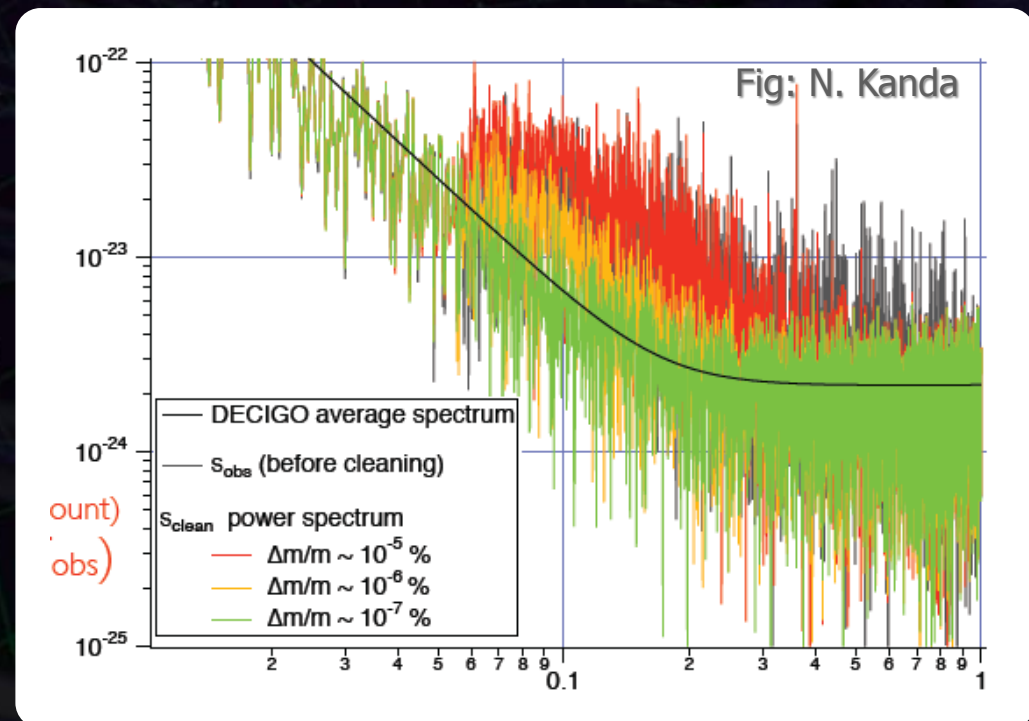
DECIGO will watch
 $\sim 10^5$ NS binaries

⇒ **Foreground for GWB**

In principle, possible
to remove them.

Require waveform

Accuracy $\Delta m/m < \sim 10^{-7}$ %



Considering “Conceptual design”

By T.Akutsu

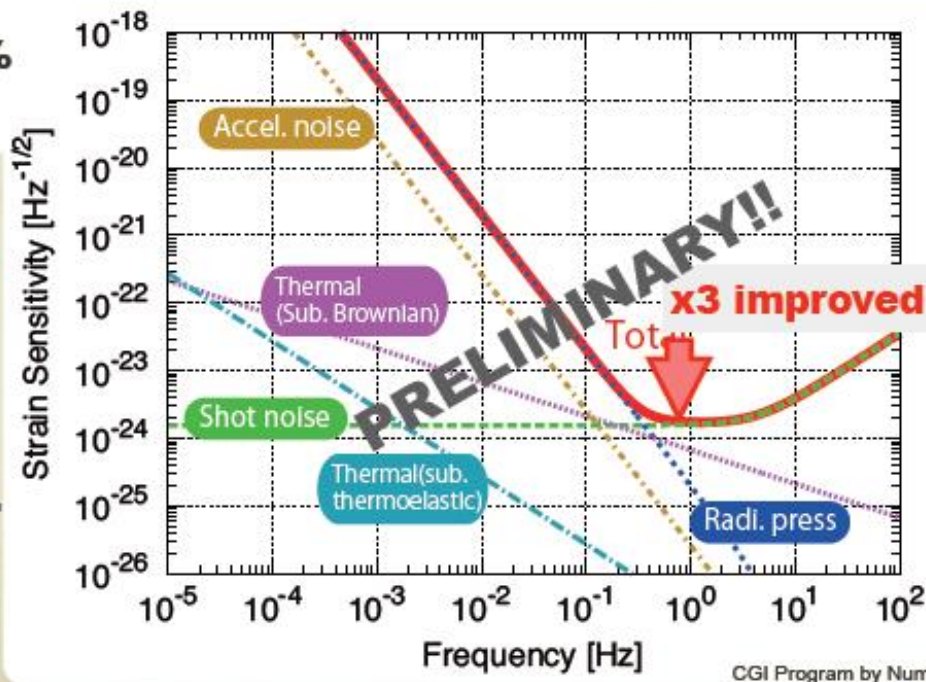
- Arm length: **1,500 km**
- Laser power: **30 W**
- Laser wavelength: **532 nm**
- Mirror diameter: **1.5 m**
- Mirror mass: **100 kg**
- Mirror reflectivity: **77.3%**
- Cavity g-param: **0.1**

Preliminary
← Parameters tuned

This is the first step to considering the **conceptual design**.

Next:

- ➔ Confirm the calculations.
- ➔ Find the realistic way to realize this!



DECIGO Pathfinder

Roadmap for DECIGO

Figure: S.Kawamura

	2014	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
Mission																				
Purpose	Interferometer in Space (Obs. Of GW and Earth Gravity)						Long-baseline Precise Formation Flight (GW Observation)						GW Astronomy and Cosmology							
Design	One Small Satellite Short FP cavity + Drag-free						FF with 3 S/C 1 IFO unit						FF with 3 S/C 3-4 IFO units							

- Key technologies for DECIGO

- (1) Precise measurement by laser interferometer.

- Operation of Fabry-Perot interferometer
in Space environment and Drag-free control.

- ⇒ Demonstration by DPF

- (2) Long-baseline formation flight.

- Realization of precise formation flight
with more than km scale

- ⇒ Demonstration by Pre-DECIGO

Technical Steps for DECIGO



	DPF target	Pre-DECIGO target	DECIGO Requirement
Space FP	First demonstration of FP cavity (30cm) in space. Disp. noise $\sim 10^{-16} \text{m/Hz}^{1/2}$, Acc. Noise $10^{-15} \text{N/Hz}^{1/2}$.	FP operation with long-base line (100km). Disp. noise $10^{-17} \text{m/Hz}^{1/2}$. Acc. noise $10^{-16} \text{N/Hz}^{1/2}$.	Disp. $3 \times 10^{-18} \text{m/Hz}^{1/2}$. Acc. $10^{-17} \text{N/Hz}^{1/2}$. Baseline length 1000km .
Stab. Laser source	Freq. stability of $0.5 \text{Hz/Hz}^{1/2}$ in space environment. Output pow. : 100mW .	Freq. stability of $0.5 \text{Hz/Hz}^{1/2}$. Output pow. : 1W .	Freq. Stab. of $0.5 \text{Hz/Hz}^{1/2}$. Output pow. : 10W .
Drag-free Control and FF	Realize all DoF drag-free control with $1 \times 10^{-9} \text{m/Hz}^{1/2}$.	All DoF DF control $1 \times 10^{-9} \text{m/Hz}^{1/2}$. Long-baseline Formation Flight 100km .	All DoF DF control $1 \times 10^{-9} \text{m/Hz}^{1/2}$. Long-baseline FF 1000km .

DECIGO Pathfinder (DPF)

First milestone mission for DECIGO

Shrink arm cavity

DECIGO 1000km \rightarrow DPF 30cm

Purpose

- FP interferometer in space
- Stabilized laser source
- Drag-free control
- Continuous data-processing



DPF satellite

DPF Payload

Size : 950mm cube
Weight : 200kg
Power : 130W
Data Rate: 800kbps
Mission thruster x10

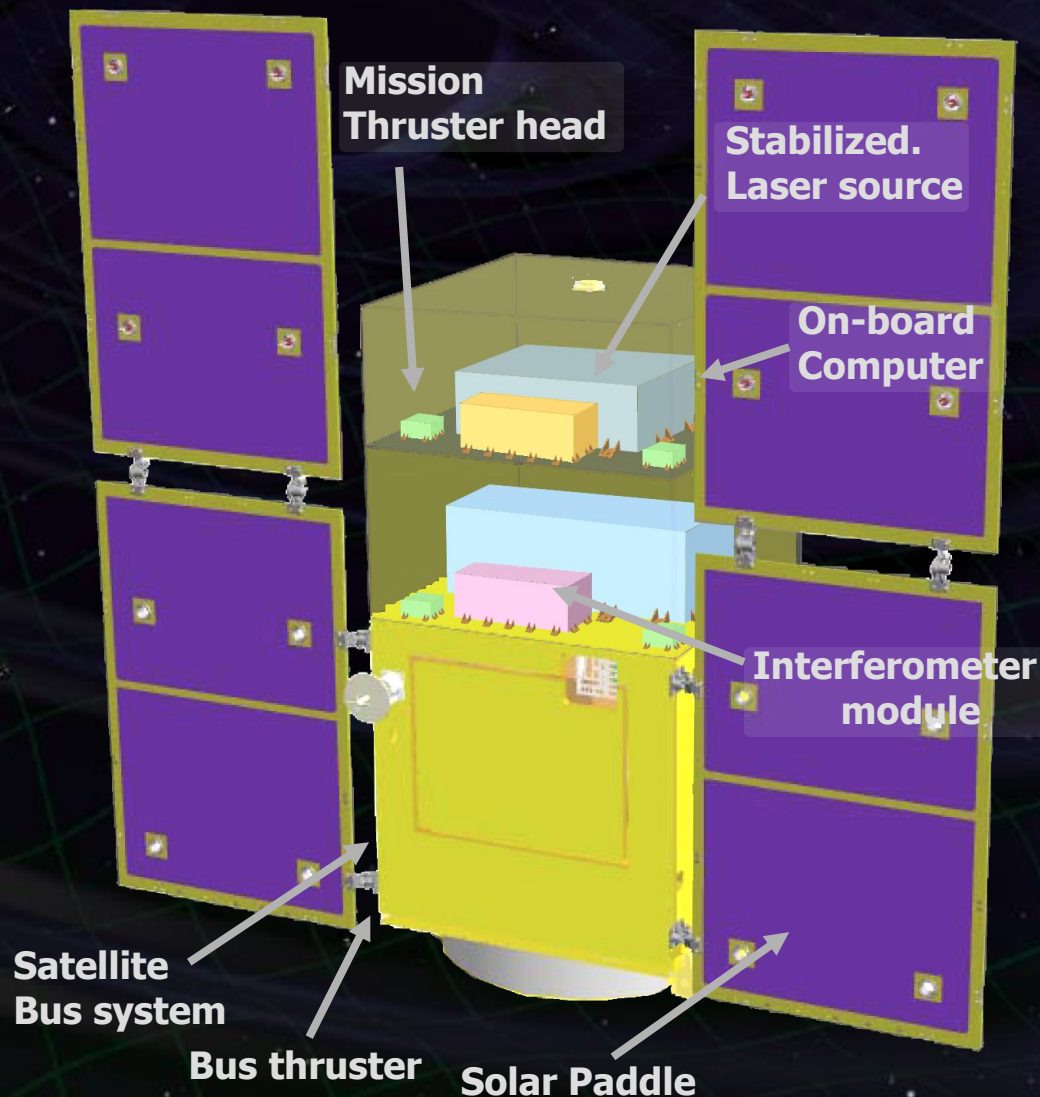
Power Supply
SpW Comm.



Satellite Bus

(‘Standard bus’ system)

Size :
950x950x1100mm
Weight : 250kg
SAP : 960W
Battery: 50AH
Downlink : 2Mbps
DR: 1GByte
3N Thrusters x 4

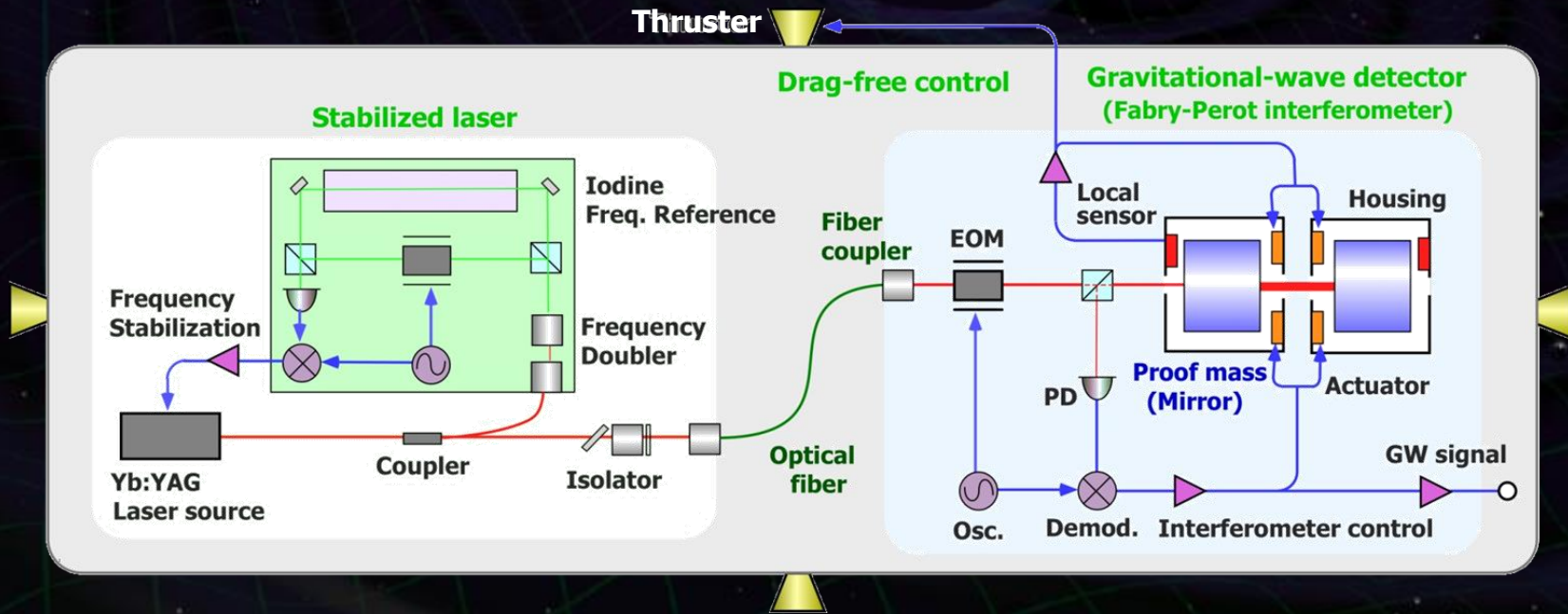


DPF mission payload

Mission weight : $\sim 200\text{kg}$
Mission space : $\sim 95 \times 95 \times 90 \text{ cm}$

Drag-free control

Local sensor signal
→ Feedback to thrusters



Laser source

Yb:YAG laser (1030nm)
Power : 25mW
Freq. stab. by Iodine abs. line

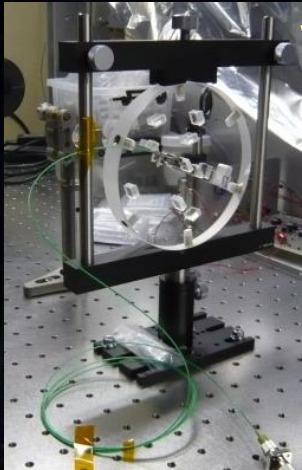
Fabry-Perot interferometer

Finesse : 100
Length : 30cm
Test mass : \sim a few kg
Signal extraction by PDH

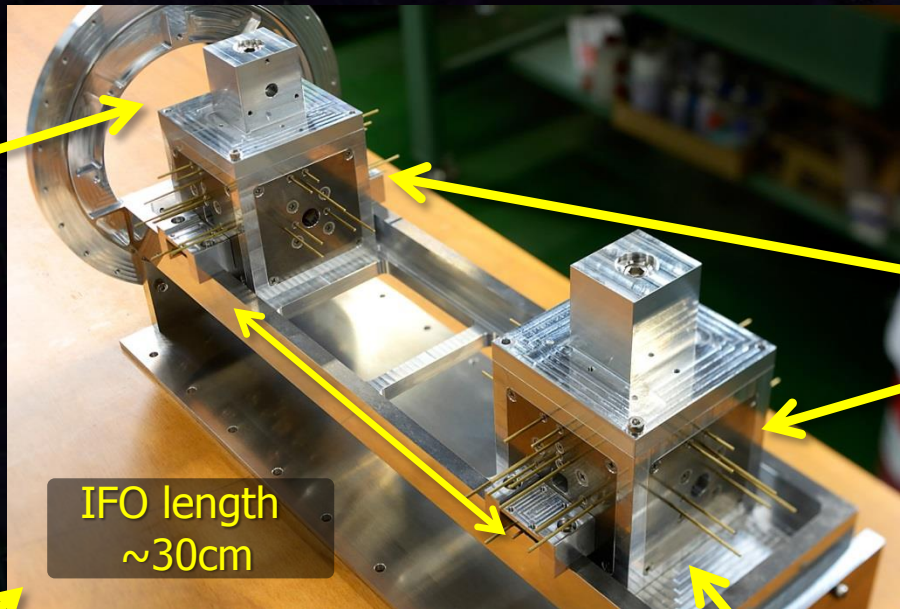
Interferometer Module

IO Optics

Monolithic opt. bench by silicate bonding

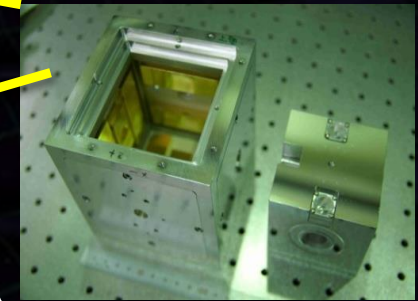


Interferometer Module



Test mass module

TM, Capacitive Sensor/Actuator, Launch lock



Quad-RFPD

Quadrant PD + Demod. circuits for length and alignment control signals



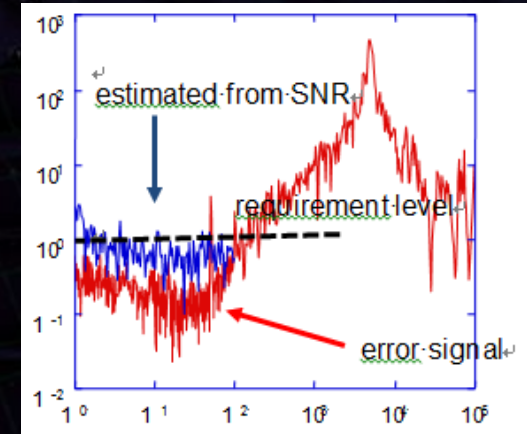
SpW signal-processing board

SpW FPGA + 16bit AD/DA

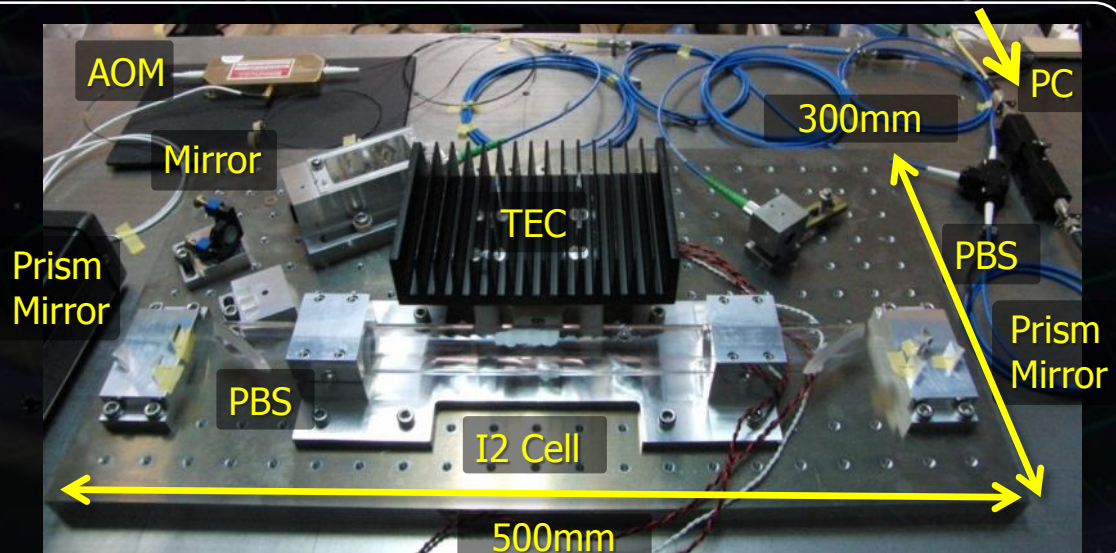
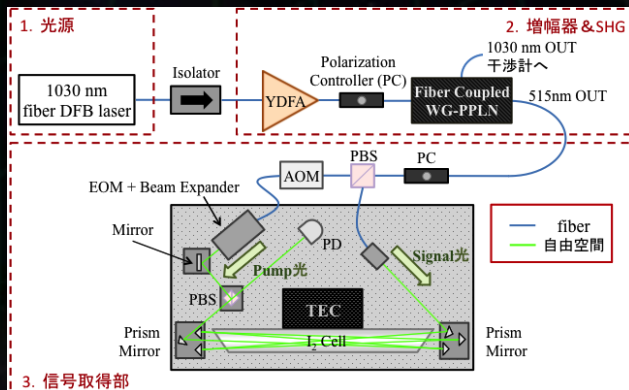


Frequency Stabilization Module

- Frequency Stabilization module BBM2 (at UEC)
 - Use absorption line of Iodine molecule.
 - Satisfy requirement ($0.5 \text{ Hz}/\text{Hz}^{1/2}$) in error-signal measurement.
 - Preparing one-more module for relative stability evaluation.



Freq. Stab module



Mission Selection Result

- AO from JAXA (December 2013)
 - for small science mission using epsilon rocket.
 - The program framework was changed:
 - ~10 M\$ payload mission → ~150 M\$ mission
 - Deadline : End of February.
 - 7 mission proposals including DPF.



DPF was dropped in the first down-selection (May)

- Started discussions on the next strategy.

Restructure of Space Program

- ISAS/JAXA decided a new plan for space science and exploration program (2014)
 - Three categories
 - * **Strategic medium-scale missions (~300 M\$)**
Hayabusa-2, ASTRO-H
 - * **Small-scale missions (100 - 150 M\$)**
AO in every two years
HISAKI, ERG, ...
 - * **Various small projects (~10 M\$/year)**
ISS missions, International collaboration,
Small rocket, Balloon, ...



'Small-scale' mission became core program in JAXA

Mission Plan by JAXA



From file submitted to the government by ISAS/JAXA

(内閣府・宇宙政策委員会・宇宙科学・探査部会 2013年9月19日)

分類	ミッション・事業名称	状況	第2期中期計画				第3期中期計画				第4期中期計画				第5期中期計画				備考		
			FY2010	FY2011	FY2012	FY2013	FY2014	FY2015	FY2016	FY2017	FY2018	FY2019	FY2020	FY2021	FY2022	FY2023	FY2024	FY2025		FY2026	FY2027
戦略的に実施する 中型計画	はやぶさ2	開発中	▲PJ準備 ▲PJ移行		▲打上		▲打上		▲小衛星到着 ▲地球帰還												
	ASTRO-H	開発中	▲PJ移行 ▲PDR ▲COR1-2		▲打上		▲打上														
	将来計画 (仮称:M1-M4) 4年に1回AO発出 開発期間6年 (5~7年)	計画中	★公募(AO) ▲PJ移行				★公募(AO) ▲PJ移行 ▲打上				★公募(AO) ▲PJ移行 ▲打上				FY2021(20-22)▲ FY2025(26-28)▲ FY2029(30-33)▲ ▲FY2033(32-34)▲						
公募型 小型計画	惑星分光衛星衛星	開発中	▲SOR/PJ移行		▲打上		▲打上														
	ジオスペース探査衛星	開発中	▲MDR/SOR ▲RSP ▲SOR ▲POR		▲打上		▲打上														
	BepiColombo	開発中	▲COR		▲打上 ▲打上		▲打上		▲水星到着												
将来計画 (仮称:S1-S7) 2年に1回AO発出 開発期間4年	計画中	★公募(AO) ▲PJ移行				★公募(AO) ▲PJ移行 ▲打上				★公募(AO) ▲PJ移行 ▲打上				FY2019▲ FY2020▲ FY2022▲ FY2024▲ FY2026▲ FY2028▲ FY2030▲							
多様な小規模 プロジェクト群	計画中	★計画決定				異種科学衛星の状況に応じて、随時AO発出・計画決定・実行する。															
基盤的 活動費	学術研究・実験等 軌道上衛星の運用 宇宙科学施設維持	継続的に 実施中	将来のミッション創出につながる学術研究・実験等の推進や衛星運用、施設維持の実施に必要な活動費。 従前より効率化努力を行って来たところ、更なる効率的な執行に努める。																		

Summary

DECIGO : Fruitful Sciences

Very beginning of the Universe

Dark energy, Dark matter

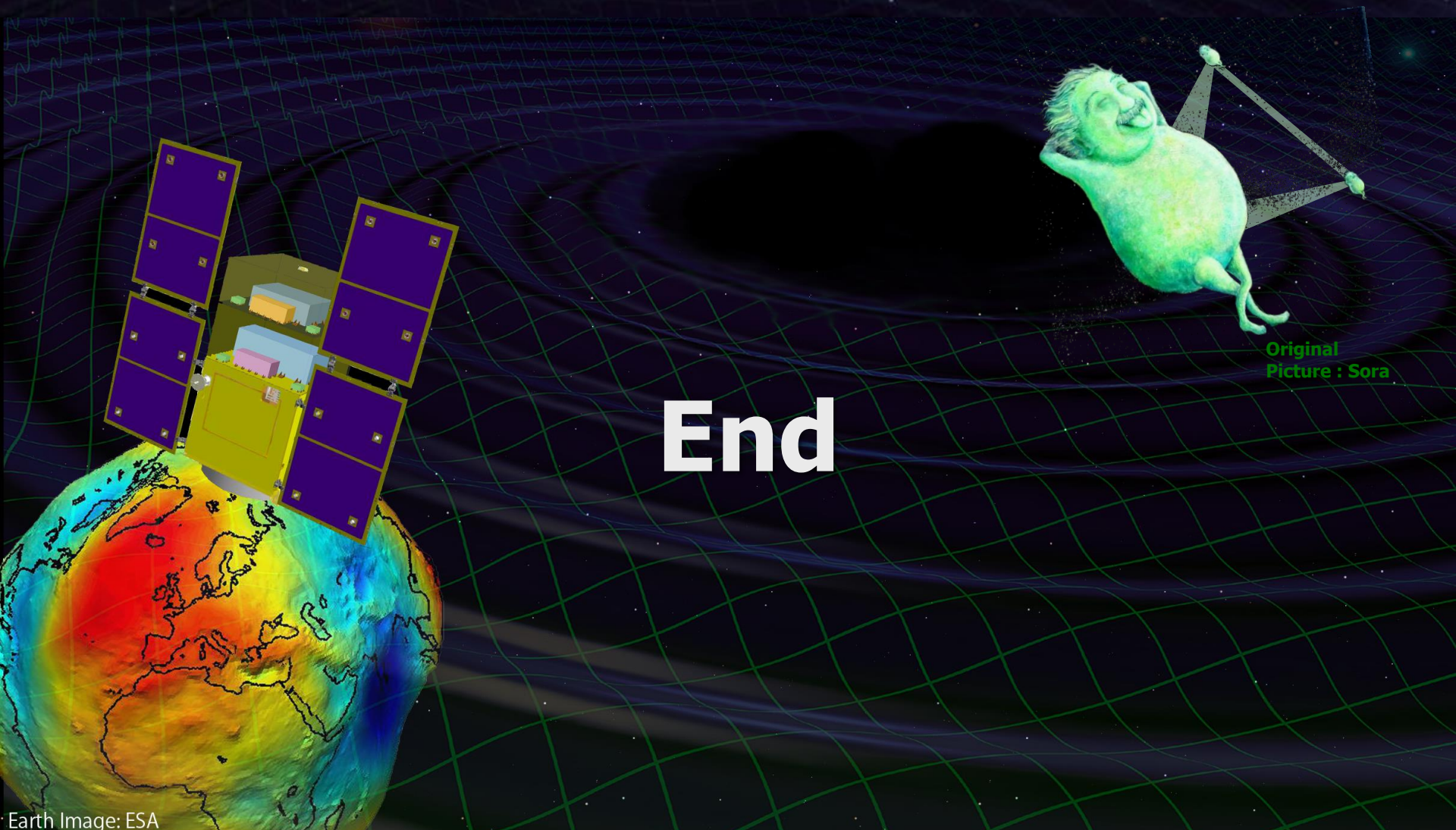
Galaxy formation

→ Will be realized at last.

DECIGO Pathfinder

Submitted mission proposal,
but failed in the selection.

→ Start discussions on the next strategy.



End

Original
Picture : Sora

Earth Image: ESA